

PŘÍKAZ K ZACHOVÁNÍ LETOVÉ ZPŮSOBILOSTI

Číslo: CAA-AD-020/1999R1

Nahrazuje CAA-AD-020/1999

Datum vydání: 19. června 2002

LETOUN - TRUP LETADLA (ATA 53) - KONTROLA

Týká se: letadel Airbus A310 a A300-600 všech verzí, vyjma verzí A300F4-605R a F4-622R, všech výrobních čísel, na kterých byla provedena AIRBUS modifikace č. 06925.

Datum účinnosti: 08. srpna 2002

Provést v termínech: Jak je popsáno v DGAC AD 1999-013-276(B) R1, od data účinnosti tohoto PZZ.

Postup provedení prací: Dle DGAC AD 1999-013-276(B) R1 (příloha tohoto PZZ).

Poznámky: Provedení tohoto PZZ musí být zapsáno do letadlové knihy. Případné dotazy týkající se tohoto PZZ adresujte na ÚCL sekce technická - Ing. Toman. Pokud to vyžaduje povaha tohoto PZZ, musí být zapracován do příslušné části dokumentace pro obsluhu, údržbu a opravy letadla. Tento PZZ byl vypracován na základě DGAC AD 1999-013-276(B) R1, který nahrazuje DGAC AD 1999-013-276(B).

Ing. Pavel MATOUŠEK
Ředitel sekce technické
Úřad pro civilní letectví

DGAC AD No.: 1999-013-276(B) R1

AIRBUS

A310 and A300-600 aircraft

Fuselage - Frame FR73A at the level of the aft pax door (ATA 53)

APPLICABILITY:

AIRBUS A310 and A300-600 aircraft, all certified models except for A300F4-605R and F4-622R models, all serial numbers, on which AIRBUS serial modification No. 06925 has been embodied.

REASONS:

During fatigue tests performed for the A330/A340 program, a crack was found on frame FR73A inner flange between beams 5 and 6.

Frames FR73A of A330/A340 aircraft are identical to the frames installed on the A310 and A300-600 aircraft on which AIRBUS serial modification No. 06925 has been embodied.

The propagation of such crack on the full width of the inner flange, and extending into frame FR73A web, could affect the structural integrity of the airframe.

The purpose of Revision 1 of this Airworthiness Directive (AD) specifies the Applicability paragraph.

COMPLIANCE:

Before accumulation of 18,000 flight cycles or within 3,000 flight cycles following the effective date of this AD at original issue, whichever occurs later, perform a High Frequency Eddy Current inspection (HFEC) of the inner flange of frame FR73A between beams 5 and 7, in accordance with the instructions of AIRBUS INDUSTRIE Service Bulletins

(SB) A310-53-2107 or A300-53-6116.

Depending on the findings of the above mentioned inspection, perform the necessary repairs before the next flight and repeat the inspections at the intervals and in accordance with the instructions defined in SB A310-53-2107 or A300-53-6116.

All findings have to be reported to AIRBUS.

REF.: AIRBUS INDUSTRIE Service Bulletins
A310-53-2107
A300-53-6116
(Any later approved revision of these SB is acceptable).

This Revision 1 replaces original AD 1999-013-276(B) issued on January 13, 1999.

EFFECTIVE DATES:

Original AD: JANUARY 23, 1999

Revision 1: JUNE 08, 2002