


EASA	AIRWORTHINESS DIRECTIVE
	<p>AD No.: 2008-0187</p> <p>Date: 10 October 2008</p> <p>Note: This Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EC) No 216/2008 on behalf of the European Community, its Member States and of the European third countries that participate in the activities of EASA under Article 66 of that Regulation.</p>
<p>This AD is issued in accordance with EC 1702/2003, Part 21A.3B. In accordance with EC 2042/2003 Annex I, Part M.A.301, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [EC 2042/2003 Annex I, Part M.A.303] or agreed with the Authority of the State of Registry [EC 216/2008, Article 14(4) exemption].</p>	
<p>Type Approval Holder's Name :</p> <p>AIRBUS SAS</p>	<p>Type/Model designation(s) :</p> <p>A310 aircraft</p>
TCDS Number :	France No 145
Foreign AD :	Not applicable
Supersedure :	This AD supersedes DGAC France AD F-2005-001, EASA approval 2004-12541.
ATA 57	Wings – Centre Wing Rear Spar Internal Angle & Tee Fitting – Inspection / Modification
Manufacturer(s):	AIRBUS (formerly AIRBUS INDUSTRIE).
Applicability:	AIRBUS A310 aircraft, all certified models, all serial numbers.
Reason:	<p>Analysis performed in the frame of the Extended Service Goal has led Airbus to modify the inspection programme [modification of thresholds, intervals and associated configurations] which is currently required by DGAC France AD F-2005-001.</p> <p>This modified inspection programme is necessary to detect and prevent damage associated with a structural fatigue phenomenon of the rear spar internal angle and the tee fitting located in the centre wing box. This condition, if not corrected, could affect the structural integrity of the centre wing box.</p> <p>For the reason stated above, this new EASA AD retains the requirements of DGAC France AD F-2005-001, which is superseded, and refers to the latest revision of Airbus Service Bulletin (SB) A310-57-2047.</p>
Effective Date:	24 October 2008

<p>Required Action(s) and Compliance Time(s):</p>	<p>Required as indicated, unless accomplished previously:</p> <p>1. Rear spar internal angle inspection</p> <p>1.1 At the threshold defined in Airbus SB A310-57-2047 Revision 07, accomplish the inspection, and apply the corrective action if necessary, of the holes in the rear spar internal angle located in the centre wing box in accordance with the instructions of Airbus SB A310-57-2047 Revision 07.</p> <p>1.2 According to the results of the previous inspection and the corrective action taken, repeat the inspection and the corrective action, if necessary, at the intervals defined in, and in accordance with the instructions of Airbus SB A310-57-2047 Revision 07.</p> <p>Aircraft which, on the effective date of this AD, have already exceeded the values of the inspection threshold or the inspection interval must be inspected at the earliest of the times specified after Table 2 of Airbus SB A310-57-2047 revision 07, to be calculated from the effective date of this AD, and without exceeding the previous values quoted in Airbus SB A310-57-2047 Revision 06.</p> <p>2. Tee fitting inspection</p> <p>2.1 At the threshold defined in Airbus SB A310-57-2047 Revision 07, accomplish the inspection and, if necessary, apply the corrective action of the tee fitting, LH side and RH side in accordance with the instructions of Airbus SB A310-57-2047 Revision 07.</p> <p>Aircraft which, on the effective date of this AD, have already exceeded the values of the inspection threshold must be inspected at the earliest of the times specified after Table 3 of Airbus SB A310-57-2047 Revision 07, to be calculated from the effective date of this AD, and without exceeding the previous values quoted in Airbus SB A310-57-2047 Revision 06.</p> <p>2.2 According to the results of the previous inspection and the corrective action taken, repeat the inspection and the corrective action, if necessary, at the intervals defined in, and in accordance with the instructions of Airbus SB A310-57-2047 Revision 07.</p> <p>3. Credit for previously accomplished tasks</p> <p>Any inspections (initial or repetitive) accomplished prior to the effective date of this AD in accordance with Airbus SB A310-57-2047 Revision 03 or 04 or 05 or 06 are acceptable for compliance with the requirements of this AD. After the effective date of this AD, repetitive inspections and corrective actions must be accomplished in accordance with Airbus SB A310-57-2047 at Revision 07.</p> <p>4. Modification of Angle and Tee Fitting</p> <p>Within 5 years after 15 January 2005 [effective date of AD F-2005-001], modify the holes in the angle and in the tee fitting in accordance with the instructions of Airbus SB A310-57-2035 Revision 10.</p> <p>This modification requirement does not apply to aircraft that have incorporated Airbus modifications 06672, 06673 and 07387, and aircraft modified in-service in accordance with Airbus SB A310-57-2035 at original issue or any later approved revisions.</p> <p>Incorporation of Airbus modifications 06672, 06673 and 07387, or accomplishment in-service of Airbus SB A310-57-2035, at original issue or any later approved revisions, does not exempt the aircraft from the inspection requirements of paragraphs 1 and 2 of this AD.</p>
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Ref. Publications:	AIRBUS SB A310-57-2047 Revision 07 AIRBUS SB A310-57-2035 Revision 10 The use of later approved revisions of these documents is acceptable for compliance with the requirements of this AD.
Remarks :	<ol style="list-style-type: none">1. If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD.2. This AD was posted on 03 September 2008 as PAD 08-101 for consultation until 24 September 2008. No comments were received during the consultation period.3. Enquiries regarding this AD should be referred to the Airworthiness Directives, Safety Management & Research Section, Certification Directorate, EASA; E-mail ADs@easa.europa.eu.4. For any question concerning the technical content of the requirements in this AD, please contact: AIRBUS SAS – EAW (Airworthiness Office, Telephone + 33 5 61 93 36 96, Fax: + 33 5 61 93 44 51).