


EASA	AIRWORTHINESS DIRECTIVE	
	<p>AD No : 2008-0018</p> <p>Date: 24 January 2008</p>	
<p>No person may operate an aircraft to which an Airworthiness Directive applies, except in accordance with the requirements of that Airworthiness Directive unless otherwise agreed with the Authority of the State of Registry.</p>		
<p>Type Approval Holder's Name : TURBOMECA</p>	<p>Type/Model designation(s) : ARRIUS 2 turboshaft engines</p>	
<p>TCDS Number: France No. M 20</p>		
<p>Foreign AD: Not applicable</p>		
<p>Supersedure: None</p>		
<p>ATA 73</p>	<p>Engine Fuel & Control – Electronic Engine Control (EEC) Unit – Identification / Modification</p>	
<p>Manufacturer(s):</p>	<p>Turbomeca</p>	
<p>Applicability:</p>	<p>ARRIUS 2B1, ARRIUS 2B1A, ARRIUS 2B2, ARRIUS 2K1 and ARRIUS 2K2 engines, all serial numbers.</p> <p>These engines are known to be installed on, but not limited to, Eurocopter Deutschland EC135 and Agusta A109 twin-engine helicopters.</p>	
<p>Reason:</p>	<p>A short circuit of some tantalum capacitors inside certain EEC units may lead to an in-flight shutdown on one of the two engines resulting from:</p> <ul style="list-style-type: none"> - direct activation of the overspeed electronic protection, - non-direct activation of the electronic overspeed protection by lowering of the threshold, - spurious activation of the starting sequence; or - loss of power control with no freeze of the fuel metering-valve. <p>For the reasons described above, this Airworthiness Directive (AD) requires the identification and modification of EEC units that have the tantalum capacitors installed that could have become brittle during their acceptance test.</p>	
<p>Effective Date:</p>	<p>07 February 2008</p>	
<p>Compliance:</p>	<p>Required as indicated, unless accomplished previously:</p> <ol style="list-style-type: none"> (1) Within the next 100 Flight Hours (FH) or 2 months, whichever occurs first after the effective date of this AD, identify the installed EEC units as listed in TURBOMECA Service Bulletin (SB) N° 319 73 2835; (2) If both EEC units installed on the helicopter's engines are affected, within 	

	<p>the next 100 FH or 2 months, whichever occurs first after the effective date of this AD, modify at least one unit in accordance with the instructions of TURBOMECA SB N° 319 73 2835;</p> <p>(3) Within 300 FH or 12 months, whichever occurs first after the effective date of this AD, modify all remaining affected EEC units in accordance with the instructions of TURBOMECA SB N° 319 73 2835;</p> <p>(4) After 07 February 2009, no person shall install a spare EEC unit with a serial number as listed in TURBOMECA SB N° 319 73 2835 on any helicopter, unless it has been modified in accordance with the instructions of TURBOMECA SB N° 319 73 2835.</p>
<p>Ref. Publications:</p>	<p>TURBOMECA Service Bulletin N° 319 73 2835, Update No.1 dated 21 December 2006.</p> <p>The use of later approved revisions of this document is acceptable for compliance with the requirements of this AD.</p>
<p>Remarks :</p>	<ol style="list-style-type: none"> 1. If requested and appropriately substantiated, EASA can accept Alternative Methods of Compliance for this AD. 2. This AD was posted on 21 December 2007 as PAD 07-229 for consultation until 18 January 2008. The Comment Response Document can be found at http://ad.easa.europa.eu/ . 3. Enquiries regarding this AD should be referred to the AD Focal Point - Certification Directorate, EASA; E-mail: ADs@easa.europa.eu . 4. For any question concerning the technical content of the requirements in this AD, please contact: ARRIUS 2 Customer Support TURBOMECA - 40220 TARNOS – FRANCE Fax: +33 5 59 74 45 15; or contact your nearest technical representative at www.turbomeca-support.com