


EASA	AIRWORTHINESS DIRECTIVE	
	<p>AD No.: 2007-0053R3</p> <p>Date: 17 December 2009</p> <p>Note: This Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EC) No 216/2008 on behalf of the European Community, its Member States and of the European third countries that participate in the activities of EASA under Article 66 of that Regulation.</p>	
<p>This AD is issued in accordance with EC 1702/2003, Part 21A.3B. In accordance with EC 2042/2003 Annex I, Part M.A.301, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [EC 2042/2003 Annex I, Part M.A.303] or agreed with the Authority of the State of Registry [EC 216/2008, Article 14(4) exemption].</p>		
<p>Type Approval Holder's Name :</p> <p>AIRBUS SAS</p>		<p>Type/Model designation(s) :</p> <p>A310 aeroplanes</p>
TCDS Number :	France No.145	
Foreign AD :	Not applicable	
Revision :	This AD revises and replaces EASA AD 2007-0053R2 dated 13 March 2009.	
ATA 53, 55, 57		Fuselage / Stabilisers / Wings – Structural Fatigue – Inspection / Modification
Manufacturer(s):	Airbus (formerly Airbus Industrie)	
Applicability:	<p>A310 aeroplanes, all certified models, all manufacturer serial numbers (MSN).</p> <p>Note: The applicability for each of the actions required by this AD is identified in each relevant paragraph of the AD Compliance Section. The "Effectivity" paragraph of each SB that is referenced in this AD may be used by operators to determine which MSN is affected.</p>	
Reason:	<p>DGAC France AD 1992-106-132(B) original issue up to revision 7 has been issued in order to mandate a set of inspections/modifications which address JAR/FAR 25-571 requirements related to damage-tolerance and fatigue evaluation of structure.</p> <p>Following the Extended Design Service Goal activities part of the Structure Task Group for the A310 program, EASA AD 2007-0053 superseded DGAC France AD F-1992-106-132R7 in order to take into account the publication of Airbus Service Bulletins (SB) A310-55-2004 at Revision 5 and Airbus SB A310-53-2074 at Revision 3, whose inspection thresholds and/or intervals had been reduced.</p> <p>Revision 1 of this AD was issued to remove the mandatory requirements related to the wings (i.e. § 1.8, 1.9, 1.10, 1.13, and 1.18) from the Compliance section, which have been transferred to EASA AD 2007-0242.</p> <p>Revision 2 of this AD has been issued to remove the mandatory requirements of paragraph 1.15, 1.16 and 1.17 which have now been transferred to EASA AD</p>	

	<p>2009-0057 (§ 1.15 and 1.17) and 2009-0058 (§ 1.16) respectively.</p> <p>Revision 3 of this AD is issued to add a Note to the Applicability and amend the Required Action(s) and Compliance Time(s) section of this AD to clarify the allowed use of the referenced SBs by operators. In addition, a note has been added to paragraph 1.7 and the notes associated to paragraphs 1.1, 1.2, 1.3, 1.4, 1.5 and 1.12 have been clarified.</p>
Effective Date:	<p>Revision 3: 01 January 2010</p> <p>Original issue, R1 and R2: 09 March 2007</p>
Required Action(s) and Compliance Time(s):	<p>Required as indicated:</p> <p>1. Unless already accomplished in accordance with any previous issues of DGAC France AD 1992-106-132(B), or with this AD at original release or revision 1 or revision 2, perform the initial inspections and/or modifications defined in the Airbus SB's listed below.</p> <p>For paragraphs 1.2, 1.3, 1.4, 1.5, 1.6 and 1.12, comply with the instructions of below mentioned Service Bulletins at the defined thresholds or within 1 000 flight cycles (FC) following 17 June 1995 [the effective date of DGAC AD F-1992-106-132 Revision 3], whichever occurs later.</p> <p>For paragraph 1.1, comply with the instructions of below mentioned Service Bulletins at the defined thresholds or within 1 000 FC following 15 June 1996 [effective date of AD F-1992-106-132 Revision 4], whichever occurs later.</p> <p>For paragraph 1.7, aeroplanes which have not been inspected in accordance with the requirements of AD F-1992-106-132 R7 must be inspected in accordance with the instructions of SB A310-55-2004 Revision 5, at the defined thresholds or within 1 500 FC following the effective date of this AD, whichever occurs later.</p> <p>1.1. Airbus SB A310-53-2014 Revision 5 and Change Notice 5B:</p> <p>Fuselage - Centre Section - Reinforce bottom joint angle fitting at FR40.</p> <p>Note: paragraph 1.1 of this AD is not applicable to aeroplanes which received application of AIRBUS modification 04944 or 04800 before initial entry into service.</p> <p>1.2. Airbus SB A310-53-2016 Revision 5:</p> <p>Fuselage - Centre Section - FR47 and FR54 area - Provide local reinforcement.</p> <p>Note: paragraph 1.2 of this AD is not applicable to aeroplanes which received total application of AIRBUS modification 04980 before initial entry into service.</p> <p>1.3. Airbus SB A310-53-2054 Revision 2:</p> <p>Fuselage - Centre Section - Inspect frame reinforcement angle run out of FR46 between STGR21 and 22 LH/RH.</p> <p>Note: paragraph 1.3 of this AD is not applicable to aeroplanes which received application of AIRBUS modification 05254, or on which Airbus SB A310-53-2019R3 has been incorporated in service.</p> <p>Note: Carrying out the repair as detailed in Airbus SB A310-53-2054 Revision 2 cancels the repetitive inspections requirements of paragraph 1.3 of this AD.</p> <p>1.4. Airbus SB A310-53-2057 Revision 1:</p> <p>Fuselage - Centre Section - Inspection of the T-section connecting FR50A to the beam between FR50 and FR51 LH/RH.</p> <p>Note: paragraph 1.4 of this AD is applicable to aeroplanes which received</p>

application of AIRBUS modification 04853 except those aeroplanes which received application of AIRBUS modification 05273 or on which Airbus SB A310-53-2011 (any issue) has been incorporated in service.

1.5. Airbus SB A310-53-2059 Revision 1:

Fuselage - Inspect the lower side panel at the lap joint with the upper side panel at FR47 and STGR22, LH/RH.

Note: paragraph 1.5 of this AD is not applicable to aeroplanes which received total application of AIRBUS modification 05997 before initial entry into service, or Airbus SB A310-53-2058 in service.

Note: Carrying out a repair detailed in Airbus SB A310-53-2059 Revision 1 cancels the repeat inspections requirements of paragraph 1.5 of this AD for the corresponding repaired area.

1.6. Airbus SB A310-55-2002 Revision 4:

Stabilizers - Horizontal Stabilizer Structure - Reinforcement of junction parts.

Note: paragraph 1.6 of this AD is applicable to aeroplanes whose MSN is listed in Airbus SB A310-55-2002 Revision 4

1.7. Airbus SB A310-55-2004 Revision 5:

Stabilizer - Horizontal Tailplane Structure - Inspection program after reinforcement as per mod. 04933 in production or by the embodiment of Airbus SB A310-55-2002 in service.

Note: paragraph 1.7 of this AD is not applicable to aeroplanes which received application of AIRBUS modification 06070 before initial entry into service.

1.8. Transferred to EASA AD 2007-0242.

1.9. Transferred to EASA AD 2007-0242.

1.10. Transferred to EASA AD 2007-0242.

1.11. Cancelled by Revision 6 of DGAC AD F-1992-106-132

1.12. Airbus SB A310-57-2039 Original Issue:

Wings - Centre wing - Inspection of front spar web vertical posts.

Note: paragraph 1.12 of this AD is not applicable to aeroplanes which received application of AIRBUS modification 04977 before initial entry into service.

1.13. Transferred to EASA AD 2007-0242.

1.14. Cancelled by Revision 7 of DGAC AD F-1992-106-132

1.15. Transferred to EASA AD 2009-0057.

1.16. Transferred to EASA AD 2009-0058.

1.17. Transferred to EASA AD 2009-0057.

1.18. Transferred to EASA AD 2007-0242.

2. Repeat the inspections defined in the SB's mentioned in paragraph 1 above, in accordance with the intervals indicated in each of the SB's when applicable.

After 09 March 2007 (the effective date of the original issue, R1 and R2 of this AD), for paragraph 1.7, accomplish the instructions of Airbus SB A310-55-2004 Revision 5 at the intervals defined therein, following the latest inspection or within the grace period defined in Airbus SB A310-55-2004 Revision 5, whichever occurs later and without exceeding the previous requirements as defined in DGAC France AD F-1992-106-

	<p>132R7.</p> <p>Note: Operators may also determine, by referring to the appropriate SB, under which condition(s) the relevant inspection program may be terminated.</p>
Ref. Publications:	<p>AIRBUS Service Bulletins: A310-53-2014 Revision 5 and change notice 5B A310-53-2016 Revision 5 A310-53-2054 Revision 2 A310-53-2057 Revision 1 A310-53-2059 Revision 1 A310-55-2002 Revision 4 A310-55-2004 Revision 3, Revision 4 or Revision 5 A310-57-2039 original issue</p> <p>The use of later approved revisions [of these documents] is acceptable for compliance with the requirements of this AD.</p>
Remarks :	<ol style="list-style-type: none"> 1. If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD. 2. The original issue of this AD was posted on 21 November 2006 as PAD 06-251 for consultation until 21 December 2006 and republished on 14 December 2006 as PAD 06-251R1, for consultation until 04 January 2007. The Comment Response Document can be found at http://ad.easa.europa.eu/. 3. Enquiries regarding this AD should be referred to the Airworthiness Directives, Safety Management & Research Section, Certification Directorate, EASA. E-mail ADs@easa.europa.eu. 4. For any question concerning the technical content of the requirements in this AD, please contact: Airbus SAS – EAW (Airworthiness Office, Telephone:+ 33 5 61 93 36 96, Fax:+ 33 5 61 93 44 51)