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PŘÍKAZ K ZACHOVÁNÍ LETOVÉ ZPŮSOBILOSTI

Číslo: 2006-16-06

Datum účinnosti: 5. září 2006

General Electric Company
motor (GE) CF6-80

Tento PZZ je vydáván pro výrobek transferovaný pod působnost EASA

Na základě rozhodnutí EASA je následující Příkaz k zachování letové způsobilosti závazný pro všechny výrobky provozované v EU na které se daný PZZ vztahuje.

Provedení PZZ, který se vztahuje podle typu a výrobního čísla na výrobek je pro provozovatele/vlastníka letadla zapsaného do leteckého rejstříku závazné. Neprovedením PZZ ve stanoveném termínu dojde ke ztrátě letové způsobilosti výrobku.

Poznámky:

- Provedení tohoto PZZ musí být zapsáno do provozní dokumentace letadla.
- Případné dotazy týkající se tohoto PZZ adresujte na ÚCL sekce technická.
- Pokud to vyžaduje povaha tohoto PZZ, musí být zpracován do příslušné části dokumentace pro obsluhu, údržbu a opravy letadla.

[Federal Register: August 18, 2006 (Volume 71, Number 160)]
[Rules and Regulations]
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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 2004-NE-05-AD; Amendment 39-14706; AD 2006-16-06]

RIN 2120-AA64

Airworthiness Directives; General Electric Company (GE) CF6-80 Series Turbofan Engines

AGENCY: Federal Aviation Administration (FAA), Department of Transportation (DOT).

ACTION: Final rule; request for comments.

SUMMARY: The FAA is superseding an existing airworthiness directive (AD) for GE CF6-80 series turbofan engines with certain stage 1 high-pressure turbine (HPT) rotor disks. That AD currently requires an initial inspection as a qualification for the mandatory rework procedures for certain disks, and repetitive inspections only for certain disks for which the rework procedures were not required. That action also requires reworking certain disks before further flight, and removes certain CF6-80E1 series disks from service. This AD requires the same actions but shortens the compliance schedule for HPT disks that have not been previously inspected using AD 2004-04-07, which this AD supersedes. This AD results from a recent report of an uncontained failure of a stage 1 HPT disk. We are issuing this AD to detect and prevent cracks in the bottoms of the dovetail slots that could propagate to failure of the disk and cause an uncontained engine failure.

DATES: Effective September 5, 2006. The Director of the Federal Register approved the incorporation by reference of certain publications listed in the regulations as of September 5, 2006. The Director of the Federal Register previously approved the incorporation by reference of certain other publications listed in the regulations as of March 12, 2004 (69 FR 8801, February 26, 2004).

We must receive any comments on this AD by October 17, 2006.

ADDRESSES: Use one of the following addresses to submit comments on this ad:

By mail: Federal Aviation Administration (FAA), New England Region, Office of the Regional Counsel, Attention: Rules Docket No. 2004-NE-05-AD, 12 New England Executive Park, Burlington, MA 01803.

By fax: (781) 238-7055.

By e-mail: 9-ane-adcomment@faa.gov.

Contact General Electric Company via Lockheed Martin Technology Services, 10525 Chester Road, Suite C, Cincinnati, Ohio 45215, telephone (513) 672-8400, fax (513) 672-8422, for the service information identified in this AD.

You may examine the AD docket, by appointment, at the FAA, New England Region, Office of the Regional Counsel, 12 New England Executive Park, Burlington, MA. You may examine the service information, by appointment, at the FAA, New England Region, Office of the Regional Counsel, 12 New England Executive Park, Burlington, MA.

FOR FURTHER INFORMATION CONTACT: James Lawrence, Aerospace Engineer, Engine Certification Office, FAA, Engine and Propeller Directorate, 12 New England Executive Park, Burlington, MA 01803; telephone: (781) 238-7176, fax: (781) 238-7199.

SUPPLEMENTARY INFORMATION: On February 13, 2004, we issued AD 2004-04-07, Amendment 39-13488 (69 FR 38; February 26, 2004). That AD requires an initial inspection as a qualification for the mandatory rework procedures for certain disks, and repetitive inspections only for certain disks for which the rework procedures were not required. That action also requires reworking certain disks before further flight. That AD was the result of the manufacturer's investigation and development of a rework procedure to chamfer the aft breakedge of the dovetail slot bottom to reduce stresses. That condition, if not corrected, could result in cracks in the bottoms of the dovetail slots that could propagate to failure of the disk and cause an uncontained engine failure.

Actions Since AD 2004-04-07 Was Issued

Since AD 2004-04-07 was issued, a CF6-80A turbofan engine, installed on a Boeing 767 airplane, experienced an uncontained stage 1 HPT disk failure on June 2, 2006. The disk failure resulted in a fire and significant damage to the airplane. The event occurred during an on-ground maintenance operation.

Relevant Service Information

We reviewed and approved the technical contents of the following GE Service Bulletins (SBs) and Alert Service Bulletin (ASB) that describe procedures for removing, inspecting, and reworking certain stage 1 HPT rotor disks:

- SB No. CF6-80E1 S/B 72-0251, dated January 22, 2004;
- SB No. CF6-80A S/B 72-0779, Revision 1, dated January 22, 2004;
- SB No. CF6-80A S/B 72-0788, Revision 3, dated July 20, 2006;
- SB No. CF6-80A S/B 72-0822, dated July 20, 2006;
- ASB No. CF6-80C2 S/B 72-A1026, Revision 2, dated January 22, 2004;
- SB No. CF6-80C2 S/B 72-1089, Revision 3, dated July 20, 2006;
- SB No. CF6-80C2 S/B 72-1217, dated July 20, 2006.

FAA's Determination and Requirements of This AD

The unsafe condition described previously is likely to exist or develop on other GE CF6-80 series turbofan engines of the same type design. This AD requires rework of the dovetail slot bottom of certain stage 1 rotor disks. The disks must pass an inspection to qualify for the rework. This AD also requires removal from service of certain disks for which the rework procedures were not previously required. This AD also tightens the compliance schedule for HPT disks that have not been previously inspected using AD 2004-04-07. Operators must use the compliance schedule carried forward from AD 2004-04-07 or the new compliance schedule below, whichever occurs first:

For stage 1 HPT rotor disks with 9,000 or more cycles-since-new (CSN) on the effective date of this AD, within 250 cycles-in-service (CIS) after the effective date of this AD, or by March 31, 2007, whichever occurs first.

For stage 1 HPT rotor disks with 6,900 or more but fewer than 9,000 CSN on the effective date of this AD, within 500 CIS after the effective date of this AD, or before accumulating 9,250 CSN, or by December 31, 2007, whichever occurs first.

For stage 1 HPT rotor disk with fewer than 6,900 CSN on the effective date of this AD, before accumulating 7,400 CSN, or by December 31, 2008, whichever occurs first.

This AD also removes from service certain CF6-80E1 series disks. You must use the service information described previously to perform the actions required by this AD.

FAA's Determination of the Effective Date

Since an unsafe condition exists that requires the immediate adoption of this AD, we have found that notice and opportunity for public comment before issuing this AD are impracticable, and that good cause exists for making this amendment effective in less than 30 days.

Comments Invited

This AD is a final rule that involves requirements affecting flight safety and was not preceded by notice and an opportunity for public comment; however, we invite you to submit any written relevant data, views, or arguments regarding this AD. Send your comments to an address listed under ADDRESSES. Include "AD Docket No. 2004-NE-05-D" in the subject line of your comments. If you want us to acknowledge receipt of your mailed comments, send us a self-addressed, stamped postcard with the docket number written on it; we will date-stamp your postcard and mail it back to you. We specifically invite comments on the overall regulatory, economic, environmental, and energy aspects of the rule that might suggest a need to modify it. If a person contacts us verbally, and that contact relates to a substantive part of this AD, we will summarize the contact and place the summary in the docket. We will consider all comments received by the closing date and may amend the AD in light of those comments.

Examining the AD Docket

You may examine the AD Docket (including any comments and service information), by appointment, between 8 a.m. and 4:30 p.m., Monday through Friday, except Federal holidays. See ADDRESSES for the location.

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. Subtitle VII, Aviation Programs, describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in subtitle VII, part A, subpart III, section 44701, "General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

Regulatory Findings

We have determined that this AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the National Government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify that the regulation:

1. Is not a "significant regulatory action" under Executive Order 12866;
2. Is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and
3. Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

We prepared a summary of the costs to comply with this AD and placed it in the AD Docket. You may get a copy of this summary by sending a request to us at the address listed under ADDRESSES. Include "AD Docket No. 2004-NE-05-AD" in your request.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, under the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. The FAA amends § 39.13 by removing Amendment 39-13488 (69 FR 8801; February 26, 2004), and by adding a new airworthiness directive, Amendment 39-14706, to read as follows:

AIRWORTHINESS DIRECTIVE

www.faa.gov/aircraft/safety/alerts/
www.gpoaccess.gov/fr/advanced.html

U.S. Department
of Transportation
**Federal Aviation
Administration**



2006-16-06 General Electric Company: Amendment 39-14706. Docket No. 2004-NE-05-AD.

Effective Date

(a) This airworthiness directive (AD) becomes effective September 5, 2006.

Affected ADs

(b) This AD supersedes AD 2004-04-07 (69 FR 8801; February 26, 2004).

Applicability

(c) This AD applies to the General Electric Company (GE) CF6-80 turbofan engine models listed in the following Table 1:

Table 1.—Applicability Models, Part Numbers, Airplanes

Models	Stage 1 High Pressure Turbine (HPT) Rotor Disk Part Numbers (P/Ns)	Engines Installed on But Not Limited To
CF6-80A, CF6-80A1, CF6-80A2, CF6-80A3.	9234M67G22/G24/G25/G26, 9362M58G02/G06/G07/G09, 9367M45G02/G04/G09.	Airbus A310 and Boeing 767 airplanes.
CF6-80C2A1, CF6-80C2A2, CF6-80C2A3, CF6-80C2A5, CF6-80C2A8, CF6-80C2A5F, CF6-80C2B1, CF6-80C2B2, CF6-80C2B4, CF6-80C2B6, CF6-80C2B1F, CF6-80C2B2F, CF6-80C2B4F, CF6-80C2B5F, CF6-80C2B6F, CF6-80C2B6FA, CF6-80C2B7F, CF6-80C2D1F.	1862M23G01, 9392M23G10/G12/G21, 1531M84G02/G06/G08/ G10/G12.	Airbus A300, A310, Boeing 747, 767, and McDonnell Douglas MD11 airplanes.
CF6-80E1A2, CF6-80E1A4.	1639M41P04.	Airbus A330 airplanes.

These engines are installed on, but not limited to, the airplanes listed in Table 1 of this AD.

Unsafe Condition

(d) This AD results from a recent report of an uncontained failure of a stage 1 HPT disk. The actions specified in this AD are intended to detect and prevent cracks in the bottoms of the dovetail slots that could propagate to failure of the disk and cause an uncontained engine failure.

Compliance

(e) You are responsible for having the actions required by this AD performed within the compliance times specified unless the actions have already been done.

CF6-80A, -80A1, -80A2, and -80A3 Engines

Stage 1 HPT Rotor Disks, P/N 9362M58G09, With Chamfered Breakedges

(f) At the next piece-part exposure, for stage 1 HPT rotor disks, P/N 9362M58G09, with serial numbers (SNs) listed in Table 2 of this AD, do the following, unless already done using superseded AD 2004-04-07:

Table 2.—SNs of CF6-80A Series Stage 1 HPT Rotor Disk P/N 9362M58G09—With Chamfered Breakedges

GWN03RD7	GWN042J3	GWN04HRD	GWN04M9K
GWN03TKG	GWN04FW2	GWN04HRE	GWN04M9L
GWN03TKH	GWN04FW3	GWN04HRF	GWN04M9M
GWN03TKJ	GWN04FW4	GWN04HRG	GWN04M9R
GWN03W3M	GWN04FW5	GWN04HRH	GWN04M9T
GWN03W3N	GWN04H0M	GWN04K8N	GWN04M9W
GWN03W3R	GWN04HRA	GWN04M9J	

(1) Visually inspect the rotor disks for the presence of a chamfer on the aft breakedges of the dovetail slot bottoms. Use paragraph 3.A. of GE Service Bulletin (SB) No. CF6-80A S/B 72-0822, dated July 20, 2006, to do the inspection.

(2) For disks that have the chamfered breakedges, re-mark, fluorescent penetrant inspect (FPI), and eddy current inspect (ECI) the rotor disk. Use paragraph 3.A.(1) of the Accomplishment Instructions of GE SB No. CF6-80A S/B 72-0822, dated July 20, 2006, to re-mark and inspect the rotor disk and remove from service as necessary.

(3) For disks that do not have the chamfered breakedges, remove the disk from service. Use paragraph 3.A(2) of the Accomplishment Instructions of GE SB No. CF6-80A S/B 72-0822, dated July 20, 2006.

Stage 1 HPT Rotor Disks, P/Ns 9234M67G22, G24, G25, G26, 9367M45G04, G09, 9362M58G02, G06, G07, and 9362M58G09 with SNs not listed in Table 2 of this AD

(g) For stage 1 HPT rotor disks, P/Ns 9234M67G22, G24, G25, G26, 9367M45G04, G09, 9362M58G02, G06, G07, and 9362M58G09 with SNs not listed in Table 2 of this AD, inspect, rework, and re-mark the disks using paragraphs 3.A.(1) through 3.A.(2) of Accomplishment Instructions of GE SB No. CF6-80A S/B 72-0788, Revision 3, dated July 20, 2006, at the following, unless already done using superseded AD 2004-04-07:

(1) For both new and used stage 1 HPT rotor disks not installed in engines, inspect, rework, re-mark, and remove from service as necessary before further flight.

(2) For stage 1 HPT rotor disks that have been inspected using any version of GE SB No. CF6-80A S/B 72-0779, inspect, rework, re-mark, and remove from service as necessary at the next Engine Shop Visit (ESV) using the compliance times in the following Table 3:

Table 3.–Compliance Times for Inspection and Rework of CF6-80A Series Stage 1 HPT Rotor Disks, P/Ns 9234M67G22, G24, G25, G26, 9367M45G04, G09, 9362M58G02, G06, G07, and 9362M58G09 With SNs Not Listed in Table 2 of This AD–Previously Inspected

Stage 1 HPT Rotor Disk Cycles-Since-Last-Inspection (CSLI) on March 12, 2004 (effective date of superseded AD 2004-04-07)	Compliance Time For Inspection and Rework
(i) More than 1,500 CSLI.	At the next ESV after March 12, 2004 (effective date of superseded AD 2004-04-07), but not to exceed 4,500 CSLI.
(ii) 1,500 CSLI or fewer.	At the next ESV after March 12, 2004 (effective date of superseded AD 2004-04-07), but not to exceed 3,500 CSLI.

(3) For stage 1 HPT rotor disks which have not been inspected using any version of GE SB No. CF6-80A S/B 72-0779, inspect, rework, re-mark, and remove from service as necessary using the following Table 4 or Table 4A compliance times, whichever occurs first:

Table 4.–Compliance Times for Inspection and Rework of CF6-80A Series Stage 1 HPT Rotor Disks, P/Ns 9234M67G22, G24, G25, G26, 9367M45G04, G09, 9362M58G02, G06, G07, and 9362M58G09 With SNs Not Listed in Table 2 of This AD–Not Previously Inspected

Stage 1 HPT Rotor Disk Cycles-Since-New (CSN) on the Effective Date of This AD	Compliance Time For Inspection and Rework
(i) 9,000 or more CSN.	Within 250 cycles-in-service (CIS) after the effective date of this AD, or by March 31, 2007, whichever occurs first.
(ii) 6,900 or more but fewer than 9,000 CSN.	Within 500 CIS after the effective date of this AD, but before accumulating 9,250 CSN, or by December 31, 2007, whichever occurs first.
(iii) Fewer than 6,900 CSN.	Before accumulating 7,400 CSN, or by December 31, 2008, whichever occurs first.

Table 4A.–Compliance Times for Inspection and Rework of CF6-80A Series Stage 1 HPT Rotor Disks, P/Ns 9234M67G22, G24, G25, G26, 9367M45G04, G09, 9362M58G02, G06, G07, and 9362M58G09 With SNs Not Listed in Table 2 of This AD–Not Previously Inspected

Stage 1 HPT Rotor Disk CSN on March 12, 2004 (effective date of superseded AD 2004-04-07)	Compliance Time For Inspection and Rework
(i) 10,000 or more CSN.	At the next ESV or within 1,000 CIS after March 12, 2004 (effective date of superseded AD 2004-04-07), whichever occurs first.

Stage 1 HPT Rotor Disk CSN on March 12, 2004 (effective date of superseded AD 2004-04-07)	Compliance Time For Inspection and Rework
(ii) 5,000 or more CSN but fewer than 10,000 CSN.	At the next ESV or within 2,400 CIS after March 12, 2004 (effective date of superseded AD 2004-04-07), whichever occurs first, but before accumulating 11,000 CSN.
(iii) Fewer than 5,000 CSN.	At the next ESV or within 3,500 CIS after March 12, 2004 (effective date of superseded AD 2004-04-07), whichever occurs first, but before accumulating 7,400 CSN.

Stage 1 HPT Rotor Disks, P/N 9367M45G02

(h) For stage 1 HPT rotor disks, P/N 9367M45G02, remove the disk from service at the following times:

(1) For stage 1 HPT rotor disks not installed in engines, remove from service before further flight.

(2) For stage 1 HPT rotor disks that have been inspected before the effective date of this AD using any version of GE SB No. CF6-80A S/B 72-0779, and had more than zero CSN at the time of that inspection, remove from service at next ESV.

(3) For stage 1 HPT rotor disks that have not been inspected, or were only inspected with zero CSN before the effective date of this AD using any version of GE SB No. CF6-80A S/B 72-0779, remove from service using the following Table 5 or Table 5A compliance times, whichever occurs first:

Table 5.—Compliance Times for Removal of CF6-80A Series Stage 1 HPT Rotor Disks, P/N 9367M45G02—Not Previously Inspected

Stage 1 HPT Rotor Disk CSN on the Effective Date of This AD	Compliance Time For Removal
(i) 9,000 or more CSN.	Within 250 CIS after the effective date of this AD, or by March 31, 2007, whichever occurs first.
(ii) 6,900 or more but fewer than 9,000 CSN.	Within 500 CIS after the effective date of this AD, but before accumulating 9,250 CSN, or by December 31, 2007, whichever occurs first.
(iii) Fewer than 6,900 CSN.	Before accumulating 7,400 CSN, or by December 31, 2008, whichever occurs first.

Table 5A.—Compliance Times for Removal of CF6-80A Series Stage 1 HPT Rotor Disks, P/N 9367M45G02—Not Previously Inspected

Stage 1 HPT rotor disk CSN on March 12, 2004 (effective date of superseded AD 2004-04-07)	Compliance Time for Removal
(i) 10,000 or more CSN.	At the next ESV or within 1,000 CIS after March 12, 2004 (effective date of superseded AD 2004-04-07), whichever occurs first.
(ii) 5,000 or more CSN but fewer than 10,000 CSN.	At the next ESV or within 2,400 CIS after March 12, 2004 (effective date of superseded AD 2004-04-07), whichever occurs first, but before accumulating 11,000 CSN.
(iii) Fewer than 5,000 CSN.	At the next ESV or within 3,500 CIS after March 12, 2004 (effective date of superseded AD 2004-04-07), whichever occurs first, but before accumulating 7,400 CSN.

CF6-80C2 Series Engines

Stage 1 HPT Rotor Disks, P/N 1531M84G10, With Chamfered Breakedges, Group 1

(i) At the next piece-part exposure, for stage 1 HPT rotor disks, P/N 1531M84G10, with SNs listed in Table 6 (Group 1) of this AD, do the following, unless already done using superseded AD 2004-04-07:

Table 6.—SNs of CF6-80C2 Series Stage 1 HPT Rotor Disks, P/N 1531M84G10, With Chamfered Breakedges, Group 1

GWN03111	GWN032G3	GWN0350M	GWN035TH
GWN03114	GWN032G4	GWN0350N	GWN035TJ
GWN031N2	GWN032G5	GWN0350P	GWN035TK
GWN031N3	GWN032G6	GWN0350R	GWN035TL
GWN031N4	GWN032G7	GWN0350T	GWN035TM
GWN031N5	GWN032G8	GWN0350W	GWN03699
GWN031N6	GWN032G9	GWN035M5	GWN0369A
GWN031N7	GWN032GE	GWN035M6	GWN0369C
GWN031N8	GWN0335P	GWN035M7	GWN0369D
GWN031N9	GWN0335R	GWN035M8	GWN0369E
GWN031NA	GWN033C5	GWN035M9	GWN0369G
GWN031NC	GWN034KR	GWN035MA	GWN0369H
GWN032G1	GWN034KT	GWN035MC	GWN0369J
GWN032G2	GWN03501	GWN035MD	GWN036JG

GWN036JH	GWN039PL	GWN03ER8	GWN03K3W
GWN036JJ	GWN039PM	GWN03ER9	GWN03K40
GWN036JK	GWN039PN	GWN03ERA	GWN03K7R
GWN036JL	GWN03A4J	GWN03FTN	GWN03KR1
GWN036JM	GWN03A4K	GWN03FTP	GWN03KR3
GWN036JN	GWN03A4L	GWN03FTR	GWN03KR4
GWN03752	GWN03A4M	GWN03FTT	GWN03KR6
GWN03753	GWN03A4N	GWN03FTW	GWN03KR7
GWN03754	GWN03A4P	GWN03FW0	GWN03KR8
GWN03755	GWN03A4R	GWN03H56	GWN03KRC
GWN03756	GWN03A4T	GWN03H57	GWN03L2D
GWN03757	GWN03A4W	GWN03H58	GWN03L2E
GWN03759	GWN03C12	GWN03HTL	GWN03L2F
GWN0375A	GWN03C13	GWN03HTM	GWN03LNF
GWN0375C	GWN03C14	GWN03HTN	GWN03LNJ
GWN0375D	GWN03CA0	GWN03HTP	GWN03LNK
GWN0375E	GWN03DC9	GWN03HTR	GWN03M88
GWN037H2	GWN03DCA	GWN03HTT	GWN03M8C
GWN03981	GWN03DCC	GWN03J8T	GWN03M8E
GWN03982	GWN03DCD	GWN03J8W	GWN03M8J
GWN03983	GWN03DCE	GWN03J91	GWN03M8K
GWN03984	GWN03DCF	GWN03J92	GWN03NHN
GWN03985	GWN03DCG	GWN03JNN	GWN03NHP
GWN03986	GWN03DCH	GWN03JNP	GWN03NHR
GWN03987	GWN03DCJ	GWN03K3C	GWN03R74
GWN03988	GWN03DCK	GWN03K3D	GWN03R76
GWN03989	GWN03DCL	GWN03K3F	GWN03R78
GWN0398A	GWN03DCM	GWN03K3G	GWN03R7E
GWN0398C	GWN03DCN	GWN03K3H	GWN03R7F
GWN039PF	GWN03DCP	GWN03K3K	GWN03R9G
GWN039PG	GWN03DCR	GWN03K3L	GWN03R9H
GWN039PH	GWN03DME	GWN03K3M	GWN03R9M
GWN039PJ	GWN03DMF	GWN03K3N	GWN03R9P
GWN039PK	GWN03ER7	GWN03K3T	GWN03R9T

Table 6.—SNs of CF6-80C2 Series Stage 1 HPT Rotor Disks, P/N 1531M84G10, With Chamfered Breakedges, Group 1 Continued

GWN03RA2	GWN0418J	GWN048CF	GWN04D52
GWN03RA3	GWN0418L	GWN048CH	GWN04D54
GWN03RA5	GWN0418N	GWN048CJ	GWN04D56
GWN03RA8	GWN0418R	GWN048CK	GWN04D57
GWN03RPA	GWN04366	GWN049GJ	GWN04D58
GWN03RPC	GWN044DP	GWN049M8	GWN04D59
GWN03RPD	GWN0454H	GWN049M9	GWN04DPW
GWN04026	GWN0454M	GWN04AER	GWN04E9K
GWN0402A	GWN0454N	GWN04ALR	GWN04E9L
GWN0402F	GWN045T0	GWN04AM1	GWN04E9M
GWN0402L	GWN045T2	GWN04CGJ	GWN04EMA
GWN040R5	GWN045T8	GWN04CGN	GWN04EMK
GWN04189	GWN045TD	GWN04CGT	GWN04EML
GWN0418A	GWN045TG	GWN04CGW	GWN04EMM
GWN0418D	GWN04722	GWN04CH3	GWN04FTL
GWN0418E	GWN04729	GWN04CH5	GWN04FTM
GWN0418F	GWN047LK	GWN04CH8	GWN04FTN
GWN0418H	GWN048CD	GWN04CH9	

(1) Visually inspect the rotor disks for the presence of a chamfer on the aft breakedges of the dovetail slot bottoms. Use paragraph 3.A. of GE SB No. CF6-80C2 S/B 72-1217, dated July 20, 2006, to do the inspection.

(2) For disks that have the chamfered breakedges, re-mark, FPI, and ECI the rotor disk. Use paragraph 3.A.(1) of the Accomplishment Instructions of GE SB No. CF6-80C2 S/B 72-1217, dated July 20, 2006, to re-mark and inspect the rotor disk, and remove from service as necessary.

(3) For disks that do not have the chamfered breakedges, remove the disk from service. Use paragraph 3.A.(4) of the Accomplishment Instructions of GE SB No. CF6-80C2 S/B 72-1217, dated July 20, 2006.

CF6-80C2 Series Engines

Stage 1 HPT Rotor Disks, P/N 1531M84G10, With Chamfered Breakedges, Group 2

(j) For stage 1 HPT rotor disks, P/N 1531M84G10, with SNs listed in Table 6A of this AD, with chamfered breakedges, (Group 2):

(1) With more than 6,900 CSN, perform paragraphs (j)(3) through (j)(5) as applicable, at the next ESV, but within 500 CIS after the effective date of this AD, unless already done using superseded AD 2004-04-07.

(2) With 6,900 CSN or fewer, perform paragraphs (j)(3) through (j)(5) as applicable, at the next ESV, but before accumulating 7,400 CSN, unless already done using superseded AD 2004-04-07.

Table 6A.—SNs of CF6-80C2 Series Stage 1 HPT Rotor Disks, P/N 1531M84G10, With Chamfered Breakedges, Group 2

GWN03J90	GWN03RA4	GWN0454F	GWN048CR
GWN03K3R	GWN03RA6	GWN0454G	GWN049GH
GWN03K6J	GWN03RA7	GWN0454J	GWN049GK
GWN03K7T	GWN03RP7	GWN0454K	GWN049JL
GWN03KR2	GWN03RP9	GWN0454L	GWN049JM
GWN03KR5	GWN03RPE	GWN045T1	GWN049M7
GWN03KRA	GWN03RPF	GWN045T3	GWN04AEP
GWN03KRD	GWN03RPG	GWN045T4	GWN04AET
GWN03M89	GWN04027	GWN045T5	GWN04ALT
GWN03M8D	GWN04028	GWN045T6	GWN04ALW
GWN03M8F	GWN04029	GWN045T7	GWN04AM0
GWN03NHT	GWN0402E	GWN045T9	GWN04AM2
GWN03R73	GWN0402G	GWN045TA	GWN04AM3
GWN03R75	GWN0402H	GWN045TC	GWN04AM4
GWN03R77	GWN0402J	GWN045TE	GWN04CGL
GWN03R79	GWN0402K	GWN045TF	GWN04CHA
GWN03R7A	GWN0402M	GWN045TH	GWN04CHC
GWN03R7C	GWN0402N	GWN046F6	GWN04D55
GWN03R7D	GWN0402P	GWN046F7	GWN04DR4
GWN03R7G	GWN0418C	GWN046F8	GWN04DR9
GWN03R7H	GWN0418G	GWN04726	GWN04DRE
GWN03R9J	GWN0418K	GWN047LG	GWN04DRJ
GWN03R9K	GWN0418M	GWN047LH	GWN04E9N
GWN03R9L	GWN0418P	GWN047LJ	GWN04EM5
GWN03R9N	GWN0418T	GWN047LL	GWN04F8N
GWN03R9R	GWN0418W	GWN048CG	GWN04F8P
GWN03R9W	GWN04190	GWN048CM	GWN04FTJ
GWN03RA0	GWN04191	GWN048CN	
GWN03RA1	GWN0454E	GWN048CP	

(3) Visually inspect the rotor disks for the presence of a chamfer on the aft breakedges of the dovetail slot bottoms. Use paragraph 3.A. of GE SB No. CF6-80C2 S/B 72-1217, dated July 20, 2006, to do the inspection.

(4) For disks that have the chamfered breakedges, re-mark, FPI, and ECI the rotor disk. Use paragraph 3.A.(2) of the Accomplishment Instructions of GE SB No. CF6-80C2 S/B 72-1217, dated July 20, 2006, to re-mark and inspect the rotor disk, and remove from service as necessary.

(5) For disks that do not have the chamfered breakedges, remove the disk from service. Use paragraph 3.A.(4) of the Accomplishment Instructions of GE SB No. CF6-80C2 S/B 72-1217, dated July 20, 2006.

CF6-80C2 Series Engines

Stage 1 HPT Rotor Disks, P/N 1531M84G12, With Chamfered Breakedges

(k) For stage 1 HPT rotor disks, P/N 1531M84G12, with SNs listed in Table 6B of this AD, with chamfered breakedges:

(1) With more than 6,900 CSN, perform paragraph (k)(3) at the next ESV, but not to exceed 500 cycles after the effective date of this AD.

(2) With 6,900 CSN or fewer, perform paragraph (k)(3) at the next ESV, but before accumulating 7,400 CSN.

Table 6B.—SNs of CF6-80C2 Series Stage 1 HPT Rotor Disks, P/N 1531M84G12, With Chamfered Breakedges

GWN04CH6

GWN04G5H

GWN04M03

(3) FPI and ECI the rotor disk. Use paragraph 3.A.(3) of the Accomplishment Instructions of GE SB No. CF6-80C2 S/B 72-1217, dated July 20, 2006, to re-mark and inspect the rotor disk, and remove from service as necessary.

Stage 1 HPT Rotor Disks, P/Ns 9392M23G10, G12, G21, 1531M84G02, G06, G08, and 1531M84G10 with SNs not listed in Table 6 and Table 6A of this AD

(l) For stage 1 HPT rotor disks, P/Ns 9392M23G10, G12, G21, 1531M84G02, G06, G08, and 1531M84G10 with SNs not listed in Table 6 and Table 6A of this AD, inspect, rework, and re-mark the disks using paragraphs 3.A.(1) through 3.A.(2) of Accomplishment Instructions of GE SB No. CF6-80C2 S/B 72-1089, Revision 3, dated July 20, 2006, at the following, unless already done using superseded AD 2004-04-07:

(1) For both new and used stage 1 HPT rotor disks not installed in engines, inspect, rework, re-mark, and remove from service as necessary before further flight.

(2) For stage 1 HPT rotor disks that have been inspected before March 12, 2004 (effective date of superseded AD 2004-04-07) using GE ASB No. CF6-80C2 S/B 72-A1024, Revision 1, dated November 3, 2000, or any version of GE ASB No. CF6-80C2 S/B 72-A1026, inspect, rework, re-mark, and remove from service as necessary using the compliance times in the following Table 7:

Table 7.—Compliance Times for Inspection and Rework of CF6-80C2 Series Stage 1 HPT Rotor Disks, P/Ns 9392M23G10, G12, G21, 1531M84G02, G06, G08, and 1531M84G10 With SNs Not Listed in Table 6 and Table 6A of This AD—Previously Inspected

Stage 1 HPT Rotor Disk Cycles-Since-Last-Inspection (CSLI) on March 12, 2004 (effective date of superseded AD 2004-04-07)	Compliance Time For Inspection and Rework
(i) More than 1,500 CSLI.	At the next ESV after March 12, 2004 (effective date of superseded AD 2004-04-07), but not to exceed 4,500 CSLI.
(ii) 1,500 CSLI or fewer.	At the next ESV after March 12, 2004 (effective date of superseded AD 2004-04-07), but not to exceed 3,500 CSLI.

(3) For stage 1 HPT rotor disks that have not been inspected before March 12, 2004 (effective date of superseded AD 2004-04-07) using GE ASB No. CF6-80C2 S/B 72-A1024, Revision 1, dated November 3, 2000, or any version of GE ASB No. CF6-80C2 S/B 72-A1026, inspect, rework, remark, and remove from service as necessary using the following Table 8 or Table 8A compliance times, whichever occurs first:

Table 8.—Compliance Times for Inspection and Rework of CF6-80C2 Series Stage 1 HPT Rotor Disks, P/Ns 9392M23G10, G12, G21, 1531M84G02, G06, G08, and 1531M84G10 With SNs Not Listed in Table 6 and Table 6A of This AD—Not Previously Inspected

Stage 1 HPT Rotor Disk Cycles-Since-New (CSN) on the Effective Date of This AD	Compliance Time For Inspection and Rework
(i) 9,000 or more CSN.	Within 250 CIS after the effective date of this AD, or by March 31, 2007, whichever occurs first.
(ii) 6,900 or more but fewer than 9,000 CSN.	Within 500 CIS after the effective date of this AD, but before accumulating 9,250 CSN, or by December 31, 2007, whichever occurs first.
(iii) Fewer than 6,900 CSN.	Before accumulating 7,400 CSN, or by December 31, 2008, whichever occurs first.

Table 8A.—Compliance Times for Inspection and Rework of CF6-80C2 Series Stage 1 HPT Rotor Disks, P/Ns 9392M23G10, G12, G21, 1531M84G02, G06, G08, and 1531M84G10 With SNs Not Listed in Table 6 and Table 6A of This AD—Not Previously Inspected

Stage 1 HPT Rotor Disk CSN on March 12, 2004 (effective date of superseded AD 2004-04-07)	Compliance Time For Inspection and Rework
(i) 10,000 or more CSN.	At the next ESV or within 1,000 CIS after March 12, 2004 (effective date of superseded AD 2004-04-07), whichever occurs first.

Stage 1 HPT Rotor Disk CSN on March 12, 2004 (effective date of superseded AD 2004-04-07)	Compliance Time For Inspection and Rework
(ii) 5,000 or more CSN but fewer than 10,000 CSN.	At the next ESV or within 2,400 CIS after March 12, 2004 (effective date of superseded AD 2004-04-07), but before accumulating 11,000 CSN.
(iii) Fewer than 5,000 CSN.	At the next ESV or within 3,500 CIS after March 12, 2004 (effective date of superseded AD 2004-04-07), whichever occurs first, but before accumulating 7,400 CSN.

Stage 1 HPT Rotor Disks, P/N 1862M23G01

(m) For stage 1 HPT rotor disk, P/N 1862M23G01, remove the disk from service at the following times:

(1) For stage 1 HPT rotor disks not installed in engines, remove from service as necessary before further flight.

(2) For stage 1 HPT rotor disks that have been inspected before March 12, 2004 (effective date of superseded AD 2004-04-07), using any version of GE ASB No. CF6-80C2 S/B 72-A1026, and had more than zero CSN at the time of that inspection, remove from service at next ESV.

(3) For stage 1 HPT rotor disks that have not been inspected, or were only inspected with zero CSN before March 12, 2004 (effective date of superseded AD 2004-04-07), using any version of GE ASB No. CF6-80C2 S/B 72-A1026, remove from service using the following Table 9 or Table 9A compliance times, whichever occurs first:

Table 9.—Compliance Times for Removal of CF6-80C2 Series Stage 1 HPT Rotor Disks, P/N 1862M23G01—Not Previously Inspected

Stage 1 HPT Rotor Disk CSN on the Effective Date of This AD	Compliance Time For Removal
(i) 9,000 or more CSN.	Within 250 CIS after the effective date of this AD, or by March 31, 2007, whichever occurs first.
(ii) 6,900 or more but fewer than 9,000 CSN.	Within 500 CIS after the effective date of this AD, but before accumulating 9,250 CSN, or by December 31, 2007, whichever occurs first.
(iii) Fewer than 6,900 CSN.	Before accumulating 7,400 CSN, or by December 31, 2008, whichever occurs first.

Table 9A.–Compliance Times for Removal of CF6-80C2 Series Stage 1 HPT Rotor Disks, P/N 1862M23G01–Not Previously Inspected

Stage 1 HPT Rotor Disk CSN on March 12, 2004 (effective date of superseded AD 2004-04-07)	Compliance Time for Removal
(i) 10,000 or more CSN.	At the next ESV or within 1,000 CIS after March 12, 2004 (effective date of superseded AD 2004-04-07), whichever occurs first.
(ii) 5,000 or more CSN but fewer than 10,000 CSN.	At the next ESV or within 2,400 CIS after March 12, 2004 (effective date of superseded AD 2004-04-07), whichever occurs first, but before accumulating 11,000 CSN.
(iii) Fewer than 5,000 CSN.	At the next ESV or within 3,500 CIS after March 12, 2004 (effective date of superseded AD 2004-04-07), whichever occurs first, but before accumulating 7,400 CSN.

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Stage 1 HPT Rotor Disks, P/N 1639M41P04

(n) For stage 1 HPT rotor disks, P/N 1639M41P04, remove the rotor disks from service using paragraphs 3.A.(1) through 3.A.(2) of Accomplishment Instructions of GE SB No. CF6-80E1 S/B 72-0251, dated January 22, 2004, at the following times:

(1) For stage 1 HPT rotor disks currently in service, remove the disk using the compliance times in the following Table 10 or Table 10A compliance times, whichever occurs first:

Table 10.–Compliance Times for Removal of CF6-80E1 Stage 1 HPT Rotor Disks, P/N 1639M41P04

Stage 1 HPT Rotor Disk CSN on the Effective Date of This AD	Compliance Time For Removal
(i) 9,000 or more CSN.	Within 250 CIS after the effective date of this AD, or by March 31, 2007, whichever occurs first.
(ii) 6,900 or more but fewer than 9,000 CSN.	Within 500 CIS after the effective date of this AD, but before accumulating 9,250 CSN, or by December 31, 2007, whichever occurs first.
(iii) Fewer than 6,900 CSN.	Before accumulating 7,400 CSN, or by December 31, 2008, whichever occurs first.

**Table 10A.—Compliance Times for Removal of CF6-80E1 Stage 1 HPT Rotor Disks,
P/N 1639M41P04**

Stage 1 HPT Rotor Disk CSN on the March 12, 2004 (effective date of superseded AD 2004-04-07)	Compliance Time for Removal
(i) More than 10,000 CSN.	At the next ESV or within 600 CIS after March 12, 2004 (effective date of superseded AD 2004-04-07), whichever occurs first.
(ii) More than 5,000 CSN but fewer than or equal to 10,000 CSN.	At the next ESV or within 2,500 CIS after March 12, 2004 (effective date of superseded AD 2004-04-07), whichever occurs first, but before accumulating 10,600 CSN.
(iii) Fewer than or equal to 5,000 CSN.	At the next ESV or within 3,500 CIS after March 12, 2004 (effective date of superseded AD 2004-04-07), whichever occurs first, but before accumulating 7,500 CSN.

(2) After March 12, 2004 (effective date of superseded AD 2004-04-07), do not install any stage 1 HPT rotor disk, P/N 1639M41P04, into any engine.

Definitions

(o) For the purpose of this AD, the following definitions apply:

(1) An engine shop visit (ESV) is when the engine is removed from an aircraft for maintenance and a major engine flange is disassembled. For stage 1 HPT rotor disks that have been inspected using any version of GE SB No. CF6-80A SB 72-0779 or any version of GE ASB No. CF6-80C2 ASB 72-A1026 or GE SB No. CF6-80C2 SB 72-A1024, Revision 1, dated November 3, 2000 or are listed in Table 6A or Table 6B, the following actions, either separately or in combination with each other, are not considered ESVs for the purpose of this AD:

- (i) The removal of the upper compressor stator case solely for airfoil maintenance.
- (ii) The module level inspection of the high-pressure compressor rotor 3-9 spool.
- (iii) The replacement of stage 5 high-pressure compressor variable stator vane bushings or lever arms.

(2) Piece-part exposure is when according to the manufacturer's engine manual or other FAA-approved engine manual the stage 1 HPT rotor disk is considered completely disassembled.

Reporting Requirements

(p) Within five calendar days of the inspection, report the results of inspections that equal or exceed the reject criteria to: Engine Certification Office, FAA, Engine and Propeller Directorate, 12 New England Executive Park, Burlington, MA 01803; telephone (781) 238-7176; fax (781) 238-7199. Reporting requirements have been approved by the Office of Management and Budget and assigned OMB control number 2120-0056. Be sure to include the following information:

- (1) Engine model in which the stage 1 HPT rotor disk was installed.
- (2) Part Number.
- (3) Serial Number.
- (4) Part CSN.

(5) Part CSLI.

(6) Date and location where inspection was done.

(q) We request that you record the inspection information and results on GE Form 1653-1, entitled CF6-80A/80C Stage 1 HPT Disk Dovetail Slot Bottom Inspection. This form is available in any version of GE SB CF6-80A S/B 72-0779, or GE ASB CF6-80C2 S/B 72-A1026. We also request that a copy of the data be sent to GE Airline Support Engineering, General Electric Aircraft Engines, Customer Support Center, 1 Neumann Way, Mail Drop RM285, Cincinnati, OH 45215.

Alternative Methods of Compliance

(r) The manager, Engine Certification Office, has the authority to approve alternative methods of compliance for this AD if requested using the procedures found in 14 CFR 39.19.

Material Incorporated by Reference

(s) You must use the service information specified in Table 11 to perform the actions required by this AD. The Director of the Federal Register previously approved the incorporation by reference of General Electric Service Bulletins No. CF6-80E1 S/B 72-0251, dated January 22, 2004 and No. CF6-80A S/B 72-0779, Revision 1, dated January 22, 2004, and Alert Service Bulletin No. CF6-80C2 S/B 72-A1026, Revision 2, dated January 22, 2004, as of March 12, 2004 (69 FR 8801, February 26, 2004). The Director of the Federal Register approved the incorporation by reference of the other documents listed in Table 11 of this AD in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. You can get a copy from General Electric Company via Lockheed Martin Technology Services, 10525 Chester Road, Suite C, Cincinnati, Ohio 45215, telephone (513) 672-8400, fax (513) 672-8422. You may review copies at the FAA, New England Region, Office of the Regional Counsel, 12 New England Executive Park, Burlington, MA; or at the Office of the Federal Register, 800 North Capitol Street, NW, Suite 700, Washington, DC. Table 11 follows:

Table 11.—Incorporation by Reference

Service Bulletin No.	Page	Revision	Date
GE SB No. CF6-80E1 S/B 72-0251 Total Pages: 4	ALL	Original	January 22, 2004
GE SB No. CF6-80A S/B 72-0779 Total Pages: 34	ALL	1	January 22, 2004
GE SB No. CF6-80A S/B 72-0788 Total Pages: 11	ALL	3	July 20, 2006
GE ASB No. CF6-80C2 S/B 72-A1026 Total Pages: 38	ALL	2	January 22, 2004
GE SB No. CF6-80C2 S/B 72-1089 Total Pages: 11	ALL	3	July 20, 2006
GE SB No. CF6-80C2 S/B 72-1217 Total Pages: 12	ALL	Original	July 20, 2006
GE SB No. CF6-80A S/B 72-0822 Total Pages: 10	ALL	Original	July 20, 2006

Related Information

(t) GE ASB No. CF6-80C2 S/B 72-A1024, Revision 1, dated November 3, 2000 also pertains to the subject of this AD.

Issued in Burlington, Massachusetts, on August 10, 2006.

Francis A. Favara,

Manager, Engine and Propeller Directorate, Aircraft Certification Service.

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