

PŘÍKAZ K ZACHOVÁNÍ LETOVÉ ZPŮSOBILOSTI

CAA-AD-063/1999

Datum vydání: 10. června 1999

LETADLO - OVLÁDÁNÍ SMĚROVÉHO KORMIDLA - KONTROLA/VÝMĚNA

Týká se: všech letadel Boeing 737 certifikovaných v kterékoliv kategorii.

Datum účinnosti: 15. července 1999

Provést v termínech: Jak je popsáno v FAA AD 99-11-05.

Postup provedení prací: Dle FAA AD 99-11-05.

Poznámky: Provedení tohoto PZZ musí být zapsáno do letadlové knihy. Případné dotazy týkající se tohoto PZZ adresujte na ÚCL technický inspektorát - Ing. Toman. Pokud to vyžaduje povaha tohoto PZZ musí být zapracován do příslušné části dokumentace pro obsluhu, údržbu a opravy letadla. Tento PZZ byl vypracován na základě FAA AD 99-11-05.

Ing. Pavel MATOUŠEK
Ředitel technického inspektorátu
Úřad pro civilní letectví

99-11-05 BOEING: Amendment 39-11175. Docket 98-NM-383-AD.

Applicability: All Model 737 series airplanes, certificated in any category.

NOTE 1: This AD applies to each airplane identified in the preceding applicability provision, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (e) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

Compliance: Required as indicated, unless accomplished previously.

To prevent failure of the secondary servo valve slide in the rudder power control unit (PCU) due to cracking of the slide, and consequent rudder hardover and reduced controllability of the airplane, accomplish the following:

Displacement Testing

(a) Perform a displacement test of the secondary slide in the dual servo valve in the rudder PCU, in accordance with Boeing Alert Service Bulletin

737-27A1221, Revision 1, dated January 28, 1999 (for Model 737-100, -200, -300, -400, and -500 series airplanes); or 737-27A1222, Revision 1, dated January 28, 1999 (for Model 737-600, -700, and -800 series airplanes); at the applicable time specified by paragraph (a)(1), (a)(2), (a)(3), or (a)(4) of this AD. Repeat the displacement test on that PCU thereafter at intervals not to exceed 24,000 flight hours.

NOTE 2: Accomplishment of the initial displacement testing required by paragraph (a) of this AD in accordance with Boeing Alert Service Bulletin 737-27A1221, dated January 14, 1999 (for Model 737-100, -200, -300, -400, and -500 series airplanes); or 737-27A1222, dated January 14, 1999 (for Model 737-600, -700, and -800, series airplanes) is acceptable only for the initial compliance requirements of this AD.

(1) For Model 737-100, -200, -300, -400, and -500 series airplanes: Conduct the displacement test within 16 months after the effective date of this AD.

(2) For airplanes equipped with a PCU having part number 65-44861-12 and having serial number (S/N) 3509A or lower: Conduct the displacement test within 16 months after the effective date of this AD.

(3) For Model 737-600, -700, and -800 series airplanes having line numbers 1 through 222 inclusive: Conduct the displacement test within 16 months after the effective date of this AD.

(4) For all other airplanes: Conduct the displacement test prior to the accumulation of 24,000 total flight hours on the PCU, or within 30 days after the effective date of this AD, whichever occurs later.

Corrective Actions

(b) If the results of the displacement test required by paragraph (a) of this AD are outside the limits specified by Boeing Alert Service Bulletin 737-27A1221, Revision 1, dated January 28, 1999 (for Model 737-100, -200, -300, -400, and -500 series airplanes), or 737-27A1222, Revision 1, dated January 28, 1999 (for Model 737-600, -700, and -800 series airplanes): Prior to further flight, accomplish the actions specified in paragraphs (b)(1) and (b)(2) of this AD.

(1) Replace the valve assembly, in accordance with the applicable alert service bulletin, with a serviceable valve assembly. And

(2) Following installation of the replacement valve assembly in accordance with paragraph (b)(1) of this AD, perform the displacement test required by paragraph (a) of this AD on that assembly, in accordance with the applicable alert service bulletin. If the test results are outside the limits specified by the applicable alert service bulletin, prior to further flight, replace the valve assembly with a serviceable valve assembly in accordance with the applicable alert service bulletin, and repeat the displacement test required by paragraph (a) of this AD on that assembly.

NOTE 3: Boeing Alert Service Bulletin 737-27A1222, Revision 1, dated January 28, 1999, refers to Parker Service Bulletin 381500-27-01, dated December 22, 1998, as an additional source of service information for accomplishment of the displacement test for Model 737-600, -700, and -800 series airplanes.

(c) As of 16 months after the effective date of this AD, no person shall install on any airplane a main rudder PCU having serial number (S/N)

3509A or lower (for Model 737-100, -200, -300, -400, and -500 series airplanes) or S/N 0299 or lower (for Model 737-600, -700, and -800 series airplanes) unless that PCU's nameplate has been vibro-engraved with the letter "C" or letters greater than "C" following the serial number. PCU nameplates that have been vibro-engraved with the letter "C" or letters greater than "C" following the serial number are considered to be in compliance with the requirements for the initial inspection of this AD.

(d)

(1) Within 30 days after accomplishing the initial displacement test required by paragraph (a) of this AD: Submit a report of the testing to the Manager, Seattle Aircraft Certification Office (ACO), FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington 98055-4056; fax (425) 227-1181. The report must include the displacement testing results (both positive and negative findings), test data for any failed valve assemblies, a description of any discrepancies if found, the part number and serial number of each rudder PCU tested, and the airplane serial number.

(2) Within 30 days after accomplishing any repetitive displacement testing required by paragraph (a) of this AD: Submit a report of any failed valve assembly to the Manager, Seattle Aircraft Certification Office (ACO), FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington 98055-4056; fax (425) 227-1181. The report must include the displacement testing results of any failed valve assembly, test data for any failed valve assemblies, a description of any discrepancies found, the part number and serial number of each rudder PCU with a failed valve assembly, and the airplane serial number.

(3) Within 30 days after accomplishing the initial displacement test required by paragraph (a) of this AD: Submit failed valve assemblies for analysis to Parker Hannifin Corporation, Chief Engineer, Customer Support Operations, 16666 Von Karman Avenue, Irvine, California 92606.

(4) Information collection requirements contained in this regulation have been approved by the Office of Management and Budget (OMB) under the provisions of the Paperwork Reduction Act of 1980 (44 U.S.C. 3501 et seq.) and have been assigned OMB Control Number 2120-0056.

Alternative Methods of Compliance

(e) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, Seattle ACO. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, Seattle ACO.

NOTE 4: Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the Seattle ACO.

Special Flight Permits

(f) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

Incorporation by Reference

(g) The actions shall be done in accordance with Boeing Alert Service Bulletin 737-27A1221, Revision 1, dated January 28, 1999, or Boeing Alert Service Bulletin 737-27A1222, Revision 1, dated January 28, 1999. This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from Boeing Commercial Airplane Group, P. O. Box 3707, Seattle, Washington 98124-2207. Copies may be inspected at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

(h) This amendment becomes effective on June 28, 1999.

FOR FURTHER INFORMATION CONTACT:

R.C. Jones, Aerospace Engineer, Systems and Equipment Branch, ANM-130S, FAA, Transport Airplane Directorate, Seattle Aircraft Certification Office, 1601 Lind Avenue, SW., Renton, Washington 98055-4056; telephone (425) 227-1118; fax (425) 227-1181.