



CIVIL AVIATION AUTHORITY
CZECH REPUBLIC
Airworthiness Division
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AIRWORTHINESS DIRECTIVE

Number: CAA-AD-T-054/2000R1
Supersedes CAA-AD-T-054/2000
Date of issue: September 24, 2004
Eurocopter
AS 355

HELICOPTER – TAIL ROTOR HEAD (ATA 05, 64) – INSPECTION

Applicability: AS 355 helicopters, versions E, F, F1, F2 and N, fitted with tail rotor pitch change rotating plates all part numbers, on which EUROCOPTER modification (MOD) 07 6554 has not been embodied.

This Airworthiness Directive (AD) does not apply to pitch change plate assembly part number 350A33-2030-00 (MOD 076550).

Effective date: Upon receipt

Compliance: Required as indicated in DGAC AD F-2000-223-059 R1.

Remarks: The compliance of this AD must be recorded in Aircraft Logbook, where applicable the requirements of this AD must be integrated into Aircraft Technical Documentation. Address inquiries concerning this AD to: Civil Aviation Authority, Airworthiness Division, Ruzyně Airport, 160 08 Prague 6, Czech Republic, tel: 420 233320922, fax: 420 220562270. This AD has been issued in accordance with LBA AD D-1994-043R1, which is approved under EASA reference No. 2004-5312.

Ing. Pavel MATOUŠEK
director

DGAC AD No.: F-2000-223-059 R1

Issue Date: September 15, 2004

This Airworthiness Directive is published by the DGAC on behalf of EASA, Airworthiness Authority of the State of Design for the affected product, part or appliance.

No person may operate an aircraft to which an Airworthiness Directive applies, except in accordance with the requirements of that Airworthiness Directive, unless otherwise agreed with the Authority of the State of Registry.

Corresponding foreign Airworthiness Directive(s): Not applicable
Airworthiness Directive(s) replaced: 2000-223-059 original issue
Person in charge of airworthiness: EUROCOPTER

Type(s): AS 355 helicopters

Type certificate(s) No. 168
TCDS No 168
ATA chapter: 05,64

Subject: Tail rotor - Bearing spacer of the tail rotor head pitch change plate

1. EFFECTIVITY:

AS 355 helicopters, versions E, F, F1, F2 and N, fitted with tail rotor pitch change rotating plates all part numbers, on which EUROCOPTER modification (MOD) 07 6554 has not been embodied.

This Airworthiness Directive (AD) does not apply to pitch change plate assembly part number 350A33-2030-00 (MOD 076550).

2. REASON:

This AD is issued to prevent deterioration and loss of the tail rotor head (TRH) pitch change control.

Revision 1 of this AD covers referenced Revision 1 of EUROCOPTER AS 355 Alert Service Bulletin (ASB) No. 05.00.33, with no change to the technical content, but reducing the effectivity defined in paragraph 1 with reference to a modification.

3. MANDATORY ACTIONS AND COMPLIANCE TIMES:

The following measures are rendered mandatory as from the effective date of the original issue of this AD:

3.1. At the latest within 10 flying hours, identify the position of the spacer and the tail rotor pitch change rotating plate in compliance with the instructions given in paragraph 2.B.1. of referenced EUROCOPTER AS 355 ASB No. 05.00.33 R1.

3.2. At each check after the last flight of the day, check in compliance with the instructions given in paragraph 2.B.2. of the referenced ASB, that the paint index marks on the tail rotor pitch change rotating plate and on the spacer are aligned.

If the paint index marks are aligned, comply with the instructions described in paragraph 2.B.3. of the referenced ASB (embodiment of MOD 07 6554) at the latest during the next "T" basic inspection.

If the paint index marks are not aligned, comply with the instructions described in paragraph 2.B.4. of the referenced ASB (embodiment of MOD 07 6554) at the latest within 25 flying hours from detection of the misalignment.

3.3. Before installing a pitch change plate assembly or a tail gear box assembly, held as spares, on an aircraft, comply with the instructions described in paragraph 2.B.3. of the referenced ASB (embodiment of MOD 07 6554).

4. REFERENCE PUBLICATION:

EUROCOPTER AS 355 Alert Service Bulletin No. 05.00.33 R1 (Any subsequent approved revision to this ASB is acceptable).

5. EFFECTIVE DATES:

Original issue: Upon receipt of telegraphic AD dated June 02, 2000

Revision 1: September 25, 2004.

6. REMARK:

For any questions concerning the technical content of the requirements in this AD, please contact:

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7. APPROVAL:

This AD Revision is approved under EASA reference No 2004-9370 dated September 07, 2004.